

CLAIM SET AS AMENDED

1. (Canceled)

2. (Canceled)

3. (Currently Amended) The blade member for an airplane according to
~~claim 1~~ claim 23, wherein a distance between outer surfaces of said first outer
skin and said second outer skin is gradually decreasing toward the trailing
edge to become approximately zero at the trailing edge.

4. (Currently Amended) The blade member for an airplane according to
~~claim 1,~~ claim 24, wherein a distance between outer surfaces of said first outer
skin and said second outer skin is gradually decreasing toward the trailing
edge to become approximately zero at the trailing edge.

5. (Currently Amended) The blade member for an airplane according to
~~claim 1~~ claim 23, wherein the wall thickness of the first outer skin includes a
central portion that is thicker relative to the leading edge and trailing edge.

6. (Currently Amended) The blade member for an airplane according to
~~claim 1, claim 23~~, wherein two reinforcing areas are provided within the outer
skin area for connecting the first outer skin to the second outer skin, said two
reinforcing areas being spaced a predetermined distance relative to each other.

7. (Currently Amended) The blade member for an airplane according to
~~claim 1 claim 23~~, wherein the blade member is constructed of an aluminum
alloy.

8. (Currently Amended) The blade member for an airplane according to
~~claim 1, claim 23~~, wherein the first outer skin is curved upwardly.

9. (Currently Amended) The blade member for an airplane according to
~~claim 1, claim 23~~, wherein said second outer skin is substantially flat.

10. (Currently Amended) The blade member for an airplane according to
~~claim 1, claim 23~~, wherein two reinforcing areas are provided within the outer
skin area for connecting the first outer skin to the second outer skin, said two
reinforcing areas being spaced a predetermined distance relative to each other
and said first outer skin being curved upwardly and includes a thickened
portion extending between the two reinforcing areas.

11-20 (Canceled)

21. (Currently Amended) The blade member for an airplane according to
~~claim 1, claim 23,~~ wherein two reinforcing areas are provided within the outer
skin area for connecting the first outer skin to the second outer skin,

wherein the at least one of wall thickness of said first outer skin and said
second outer skin changes in a cord direction between a first of said two
reinforcing areas and a second of said two reinforcing areas, with a portion
adjacent to the first of said two reinforcing area being thicker than a portion
adjacent to the second of the two reinforcing area.

22. (Currently Amended) A blade member for an airplane, which
constitutes at least a portion of a rotor blade of the airplane, said blade
member comprising:

an outer skin area surrounded by a first outer skin, a second outer skin,
a leading edge and a trailing edge each having a predetermined wall thickness;
and

at least one reinforcing area extending in a span direction within the
outer skin area and connected to the first outer skin and the second outer
skin;

wherein a wall thickness of said second outer skin changes in a cord
direction between said at least one reinforcing area and the leading edge, with

a portion of the second outer skin extending in a rearward direction from said at least one reinforcing area being thicker than a portion extending in a rearward direction from the at least one reinforcing area, and

wherein said outer skin area including said first outer skin, said second outer skin, said leading edge, said trailing edge, and said at least one reinforcing area are integrally formed from a single block.

23. (New) A blade member for an airplane, which constitutes at least a portion of a rotor blade of the airplane, said blade member comprising:

an outer skin area surrounded by a first outer skin, a second outer skin, a leading edge and a trailing edge each having a predetermined wall thickness; and

at least one reinforcing area extending in a span direction within the outer skin area and connected to the first outer skin and the second outer skin;

wherein said outer skin area including said first outer skin, said second outer skin, said leading edge and said trailing edge and said reinforcing area are integrally formed from a single block by wire electrical discharge-machining.

24. (New) A blade member for an airplane, which constitutes at least a portion of a rotor blade of the airplane, said blade member comprising:

an outer skin area surrounded by a first outer skin, a second outer skin, a leading edge and a trailing edge each having a predetermined wall thickness; and

at least one reinforcing area extending in a span direction within the outer skin area and connected to the first outer skin and the second outer skin;

wherein said outer skin area including said first outer skin, said second outer skin, said leading edge and said trailing edge and said reinforcing area are integrally formed from a single block by wire electrical discharge-machining, and

wherein at least one of wall thickness of said first outer skin and said second outer skin changes in a cord direction.

25. (New) A blade member for an airplane, which constitutes at least a portion of a rotor blade of the airplane and has an asymmetrical cross-sectional shape as viewed in a span direction, said blade member comprising:

an outer skin area elongated in the span direction and surrounded by a first outer skin, a second outer skin, a leading edge and a trailing edge each having a predetermined wall thickness; and

at least one reinforcing area extending in the span direction within the outer skin area and connected to the first outer skin and the second outer skin;

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wherein said outer skin area including said first outer skin, said second outer skin, said leading edge and said trailing edge and said reinforcing area are integrally formed from a single block by wire electrical discharge-machining.